

Patent claims

1. A rotor blade (16) for a turbomachine,  
in particular for a gas turbine through which a working fluid  
5 (20) flows,  
having a rotor blade root (33), a platform region (21)  
extending transversely with respect thereto, and a curved main  
blade profile (24),  
which extends from a leading edge (25), onto which the working  
10 fluid (20) can flow, to a trailing edge (26), between which are  
formed a suction side (27) and a pressure side (28) of the main  
blade profile (24),  
and having a relief slot (30) in the main blade profile (24),  
characterized in that  
15 the relief slot (30) is arranged in the region of the trailing  
edge (26), at that end of the main blade profile (24) which  
faces the platform region (29), extends through the main blade  
profile (24) from the suction side (27) to the pressure side  
(28) and is oriented substantially transversely to the  
20 direction of flow of the working fluid (20).

2. The rotor blade (16) as claimed in claim 1, characterized  
in that the relief slot (30) is provided at a distance B, which  
amounts to at least 90% of a chord length A of a profile chord  
25 (S) measured between the leading edge (25) and the trailing  
edge (26), from the leading edge (25) of the rotor blade (16).

3. The rotor blade (16) as claimed in claim 1 or 2,  
characterized in that the relief slot (39) has a length L which  
30 is in the range from 5% to 10% of the blade height H of the  
main blade profile (24).

4. The rotor blade (16) as claimed in claim 1, 2 or 3,  
characterized in that that end of the relief slot (30) which  
faces the platform region (21) is at a distance Z of from 5% to  
10% of the blade height H of the main blade profile (24) from  
5 the platform region (21).

5. The rotor blade (16) as claimed in one of the preceding  
claims, characterized in that the relief slot (30) is rounded  
at its ends.

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6. The rotor blade (16) as claimed in one of the preceding  
claims, characterized in that the relief slot (30) is filled  
with a filler material (33) which has a coefficient of thermal  
expansion which is less than or equal to that of the blade  
15 material (34).

7. The rotor blade (16) as claimed in one of the preceding  
claims, characterized in that the filler material (33) is a  
solder.

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8. The rotor blade (16) as claimed in one of the preceding  
claims, characterized in that the rotor blade (16) is single-  
crystalline or directionally solidified.

25 9. A gas turbine (1) having the rotor blade (16) as claimed  
in one of the preceding claims.